

**AMENDMENTS TO THE CLAIMS:**

This listing of claims will replace all prior versions, and listings, of claims in the application:

**Listing of Claims:**

Claim 1 (Canceled).

Claim 2 (Currently Amended):      The process according to claim + 15, wherein the process pressure is adjusted to between 0.6 bar absolute and 1.25 bar absolute and the process temperature is adjusted to between 1100°C and 1200°C.

Claim 3 (Currently Amended):      The process according to claim + 15, wherein the silicon carbide precursor is selected from chloro(alkyl)silanes.

Claim 4 (Canceled).

Claim 5 (Canceled).

Claim 6 (Canceled).

Claim 7 (Currently Amended):      The process according to claim + 15, wherein the heat-resistant material with a large surface is a carbon fiber felt.

Claim 8 (Currently Amended):      The process according to claim + 15, wherein carbon fiber scrims are used.

Claim 9 (Currently Amended):      The process according to claim + 15, wherein a residual porosity is adjusted to be between from 12 to 14 percent of volume.

Claim 10 (Currently Amended):      The process according to claim + 15, wherein the carbon fiber preforms or silicon carbide fiber preforms are generated in that fiber layers are first constructed, the fiber layers are fixed one above the other at a distance from one

another by binders, possibly accompanied by simultaneous molding and stabilization of the preform approximating the desired end product.

Claim 11 (Previously Presented): The process according to claim 10, wherein the fiber layers are generated with a 0° or 90° laying angle to a main axis of the preform.

Claim 12 (Original): The process according to claim 10, wherein the binder is selected from organic and/or silicon-organic polymer resins.

Claim 13 (Currently Amended): Structural component parts, particularly for aircraft and spacecraft, produced according to the process of claim 1 ± 15.

Claim 14 (Currently Amended): Structural component parts for control flaps, leading edges of blades, nose cones, control rudders or heat protection paneling for orbital gliders and hypersonic aircraft produced according to the process of claim 1 ± 15.

Claim 15 (New): In a process for producing a high temperature stable fiber composite ceramic by chemical vapor infiltration (CVI) with a methyltrichlorosilane (MTS) in hydrogen (H<sub>2</sub>) on fiber scrims of carbon fiber preforms or silicon carbide fiber preforms, wherein the partial pressure ratio of hydrogen to methyltrichlorosilane is adjusted between 4 and 8, the process further comprises:

adjusting the process pressure to  $\geq 0.6$  bar absolute;

adjusting the process temperature to  $\geq 1100^{\circ}\text{C}$ ; and

arranging a heat-resistant material with a large surface between a gas feed in the reaction space and between the fiber scrims of carbon fiber preforms or silicon carbide fiber preforms to be infiltrated for preconditioning the process gas.